European Union Aviation Safety Agency

Airworthiness Directive

AD No.: 2026-0052R2

Issued: 24 September 2026

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:
AIRBUS HELICOPTERS

Type/Model designation(s):
AS 350 and EC 130 helicopters

Effective Date: Revision 02: 01 October 2026
Revision 01: 08 July 2026
Original Issue: 26 March 2026

TCDS Number(s): EASA.R.008

Foreign AD: Not applicable

Revision: This AD revises EASA AD 2026-0052R1 dated 08 July 2026.

ATA 31 – Instruments – Fuse Cover Attachment of Circuit Breaker Panel – Modification

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Manufacturer(s):
Airbus Helicopters (AH), formerly Eurocopter, Eurocopter France, Aerospatiale

Applicability:
AS 350 B2, AS 350 B3 helicopters, all serial numbers (s/n) on which modification (mod) 073273 and mod 073274 have been embodied, except those on which mod 074280 has been embodied; and EC 130 B4 helicopters, all s/n on which mod 073245 has been embodied.

Definitions:
For the purpose of this AD, the following definitions apply:

The ASB: AH Alert Service Bulletin (ASB) ASB AS350-31-10-0002 Issue 002 or ASB EC130-31-10-0001, as applicable.

Affected part: Fuse cover, having Part Number (P/N) 350A76-2395-20, P/N 350A76-2395-21 or P/N 350A76-2395-23, which is part of the control unit having a P/N as listed in Appendix 1 of this AD, except those covers which have been modified and reidentified in accordance with the instructions of the ASB.

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Reason:

An occurrence of partial blocking of a pilot pedal was reported. Further investigation indicated that this event was caused by an interference between the left pedal of the pedal block and the partially detached fuse cover of the circuit breaker panel.

This condition, if not corrected, could result in reduced yaw control of the helicopter.

To address this potential unsafe condition, AH published the original issue of the ASB, as defined in this AD, to provide instructions for modification of the affected part (as defined in this AD). Consequently, EASA published AD 2026-0052 to require modification of the affected part and to prohibit installation of a pre-mod affected part.

After that AD was issued, it was identified that for AS 350 helicopters the modification could not be accomplished as described in the ASB at original issue, and AH confirmed that, according to reported data, no AS 350 helicopter has been modified.

It was also determined that the compliance time (CT) for the modification could be extended for AS 350 helicopters. Therefore, AH updated the design of the modification for the AS 350 and published Issue 02 of ASB AS350-31-10-0002, providing amended modification instructions and introducing an extended CT for the AS 350.

Consequently, EASA AD 2026-0052 was revised to amend the definition of the ASB, now referring to Issue 02 of ASB AS350-31-10-0002, and to introduce an extended CT for the modification of AS 350 helicopters. This AD incorporates also administrative and editorial amendments to improve the readability and to meet the latest standards for EASA ADs.

Since AD 2026-0052R1 was issued, for EC 130 B4 helicopters, it was determined that the compliance time for accomplishment of the modification required by this AD can be extended. Consequently, AH published ASB EC130-31-10-0001 issue 002 to extend the CT.

This AD is revised accordingly.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the actions required by this AD have been already accomplished:

Modification:

(1) Within the compliance time as identified in Table 1 of this AD, as applicable, modify and reidentify the affected part in accordance with the instructions of the ASB.

Table 1 – Compliance Time

Helicopter Model	Compliance Time
EC 130 B4	Within 660 flight hours (FH) or 9 months, whichever occurs first after 26 March 2026 [the effective date of the original issue of this AD]

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AS 350 B2 and AS 350 B3	Within 660 FH or 9 months, whichever occurs first after 26 March 2026 [the effective date of the original issue of this AD]

Part(s) Installation:

(2) After modification of a helicopter, as required by paragraph (1) of this AD, do not install an affected part on that helicopter.

Ref. Publications:

AH ASB AS350-31-10-0002 Issue 002 dated 07 July 2026.

AH ASB EC130-31-10-0001 original issue dated 11 February 2026 or Issue 002 dated 22 September 2026.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.

2. The original issue of this AD was posted on 20 February 2026 as PAD 26-031 for consultation until 06 March 2026. No comments were received during the consultation period.

3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.

4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the EU aviation safety reporting system. This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.

5. For any question concerning the technical content of the requirements in this AD, please contact: Airbus Helicopters Customer Support, Telephone +33 (0)4.42.85.97.89, Fax + 33 (0)4.42.85.99.66, E-mail: Airframe.Technical-Support@airbus.com, Technical Request Management: TechnicalSupport.Helicopters@airbus.com.

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Appendix 1

Control Unit Affected P/N

Helicopter Model	Manufacturer P/N	P/N
AS 350 B2 AS 350 B3	052706AA	704A46580115
	052706B or 052706B-A	704A46580118
	074102AA	704A46580122
	052706CA	704A46580123
	101801AA	704A46580129
	104101AA	704A46580130
	304-2613-00	704A46580140
	304-2614-00	704A46580141
	304-2615-00	704A46580142
	304-2616-00	704A46580143
	304-2617-00	704A46580144
EC 130 B4	024803AA	704A46580112
	024803B or 024803B-A	704A46580119
	074801A or 074801AA	704A46580124
	304-2612-00	704A46580139

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